IN THE SPECIFICATION

Please amend the specification as follows:

1. Insert the following heading at page 1, before line 1:

--FIELD OF THE INVENTION--

2. Insert the following heading at page 1, line 10:

--BACKGROUND OF THE RELATED ART--

3. Insert the following heading at page 1, line 32:

--SUMMARY OF THE INVENTION--

4. Insert the following heading at page 6, line 14:

--BRIEF DESCRIPTION OF THE DRAWINGS--

5. Insert the following heading at page 6, line 30:

--DETAILED DESCRIPTION OF THE INVENTION--

6. Amend the paragraph on page 10, lines 30-35, as follows:

According to the invention, these particular conditions correspond to the following conditions:

- the difference DVc is greater than a predetermined value, for example 0.5 knots; and

- the actual speed Vc does not diverge significantly from the preset speed ~~Vcgt~~ Vctgt.

7. Amend the paragraph on page 11, lines 4-25, as follows:

According to the invention, said second unit 9 furthermore comprises a subsystem 24 which includes:

- a calculation means 25 for calculating a second difference Δ2 between said intermediate term ΔN1 and a previously recorded corrector term ΔN1mem;

- a comparator 26 for comparing this second difference Δ2 with a predetermined threshold value S;

- a means of selection 27 for selecting as corrector term ΔN1f:

  . said intermediate term ΔN1, if said second difference Δ2 is greater than said threshold value S; and

  . said previously recorded corrector term ΔN1mem, if said second difference Δ2 is less than or equal to said threshold value S,

said means of selection 27 transmitting the corrector term ΔN1f thus selected to the summator 10, preferably after filtering by a filtering means 28; and

- a memory 29 for recording the selected corrector term, the value of which will be used subsequently by the comparator ~~25~~ 26.

8. Amend the paragraph beginning on page 12, line 23, and ending on page 13, line 5, as follows:

By way of illustration, this equilibrium term N1eq can be calculated on the basis of the following equations:

$$N1eq = N1R \ * \ \sqrt{Tt / 288.15}$$

with
$$\begin{cases} N1R = f_1\left([[FMR]]\underline{FNR}, M\right) \\ FNR = Fn * 101325 / Pt \\ Fn = m * g * \left(\sin\gamma + (Cx / Cz) * \cos\gamma\right) \end{cases}$$

and
$$\begin{cases} Cx = f\left(Cz^2, M\right) \\ Cz = \left(m * g * \cos\gamma\right) / \left(0.7 * Ps * Sr * M^2\right) \end{cases}$$

in which the following parameters are used:

- Fn : the thrust of the engine 2 (N);
- m : the weight of the aircraft (kg);
- g : the acceleration due to gravity (≈ 9.81 m/s^2);
- γ : the slope of the aircraft (rd);
- M : the Mach number;
- Ps : the static pressure (Pa);
- Sr : a reference area (m^2);

- Cx : the coefficient of drag;

- Cz : the coefficient of lift;

- Tt : the total temperature (degrees Kelvin); and

- Pt : the total pressure (Pa).